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The following are solutions to the problems found in Chapter 1 of John D. Anderson's 2004 book Modern Compressible Flow. At the nose of a missile in flight, the pressure and temperature are 5.6 atm and 850 °R, respectively. Calculate the density and specific volume.

Modern Compressible Flow Solutions Chapter 1 | Aero ...

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Comment on Flow through Multiple Shock Systems Edit. In example 4.12 (p. 155, Anderson), one case uses two oblique shocks (one incident and one reflected) to change the Mach number from an upstream value, M 1, of 2.8 to a downstream Mach number, M 3, of 1.45.

Modern Compressible Flow Chapter 4 Notes | Aero ...

The inside surface of the tube is assumed a slip surface. Since the time span of the unsteady flow is short, the waves never reach the end walls, and so, conditions at these boundaries are held fixed. Comparison Data. The classical shock tube solution provides data for comparison. The text by Anderson discusses this solution.

Shock Tube - NASA

Anderson's book provides the most accessible approach to compressible flow for Mechanical and Aerospace Engineering students and professionals. In keeping with previous versions, the 3rd edition uses numerous historical vignettes that show the evolution of the field.New pedagogical features...

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Modern Compressible Flow with Historical Perspective, Edition 3 (but older versions are acceptable too) John D. Anderson, McGraw-Hill, Inc., New York, 2003. Other notes and articles will be provided by the instructor. Comments: Here is a link to a website that carries this book for a very reasonable price: Buy online .

Textbook, ME 420

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